



Planeweighs Limited

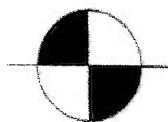
Aircraft weighing & technical services

Unit 14, Oxwich Court,
Fendrod Business Park,
Valley Way,
Swansea SA6 8RA.
Tel: (01792) 310566
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WEIGHING REPORT				No. 12305	
Type	NANCHANG CJ6A		Registration	G-CGFS	
Serial No.	4532008		Place	Seething	
Date	18-Sep-09		Weighted by	P.A. Saillard	
Equipment	1xIR-3.5 & 2xIR-10 SCALES		Calibration due	18-Feb-10	
<p>The diagram illustrates the aircraft weighing process. A vertical line represents the aircraft's centerline. A vertical line to the left is labeled 'Datum'. A horizontal double-headed arrow labeled 'x' indicates the distance from the Datum to the 'Nose wheel' (labeled '1'). A horizontal double-headed arrow labeled 'y' indicates the distance from the 'Nose wheel' to the 'Main wheels' (labeled '2'). A third point, labeled '3', is shown above the main wheels, with a vertical line connecting it to the main wheels.</p>					
Position	Ser no.	Weight (each reaction) kg	Weight (totals)	Arm (mm)	Moment (kg/mm)
1	1-003	290	290	(x) 942	273180
2	2-010	440		(y)	
3	1-010	449	889	3183	2829687
AS WEIGHED			1179	2632	3102867
Total subtractions (Column 1, see over)			110		344960
Total additions (Column 2, see over)			0		0
AIRCRAFT WEIGHT			1069	2580	2757907
Remarks: The Aircraft Weight with the fuel tanks empty (but including unusable fuel), engine oil full systems primed and equipped as per The Owner's check list dated 18-Sep-09 is 1069 kg The Centre of Gravity (Arm) is 2580 mm Aft of the Datum Procedure for Aircraft weighing is in accordance with the Aircraft Flight Manual.					
Certified that the above mentioned aircraft has been weighed in accordance with the terms of the order applicable thereto and unless otherwise stated above conforms fully to the standards/specifications quoted hereon and the requirements of the CAA.					
				Signed: For and on behalf of PLANEWEIGHS LIMITED CAA Approval no. AI/8538/79 ISO 9001-2000 Certificate No. GB93/2024	

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Column 1 - Subtractions Items weighed but not part of Basic Weight	Weight kg	Arm mm	Moment kg/mm
Fuel	110	3136	344960
Total	110		344960
Column 2 - Additions Items not in aircraft when weighed			
Nil			
Total	0		0
Notes			



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WEIGHT AND CENTRE OF GRAVITY SCHEDULE

REF NUMBER: PWL 12305

AIRCRAFT DESIGNATION: CJ6A Chujiao

NATIONALITY AND REGISTRATION MARK: G-CGFS

CONSTRUCTOR: NANCHANG

CONSTRUCTOR'S SERIAL NUMBER: 4532008

CENTRE OF GRAVITY LIMITS:

Landing Gear Extended: 2691 to 2803 mm Aft of Datum
Landing Gear Retracted: 2691 to 2815 mm Aft of Datum

MAX AUTHORISED WEIGHT: 1400 kg

PART A. BASIC WEIGHT

The Basic Weight of the aircraft as calculated from Weighing Report Planeweighs no. 12305 dated the 18-Sep-09 is: **1069** kg.

The Centre of Gravity of the aircraft in the same condition, at this weight and with the landing gear extended is: **2579.89** mm Aft of the Datum.

The total moment about the datum in this condition is: **2757907** kg/mm

NOTE: The datum is the one to which the limits in the Certificate of Airworthiness of Flight Manual refer and is defined as: **Front of propeller shaft**

The Basic Weight includes the weight of unusable fuel and the weight of the following items, which comprise the Basic Equipment:

ITEM

Two Seats c/w Harnesses
Flight & Engine Instrumentation
Funkwerk ATR500 Transceiver
Sigtronic SPA400 Intercomm
Battery

SMOKE OIL TANK AND INSTAL.

PART B. VARIABLE LOAD

The weight and lever arms of the variable load are shown below. The variable load depends on the equipment carried for the particular role.

<u>ITEM</u>	<u>WEIGHT (kg)</u>	<u>LEVER ARM (mm)</u>	<u>MOMENT (kg/mm)</u>
Pilot fwd	Use actual	2586	As calculated
Pilot aft	Use actual	3772	As calculated

NOTE: The actual weight of the pilot must be used for aircraft not exceeding 12500 lbs and with less than 12 seating capacity.

PART C. LOADING INFORMATION (DISPOSABLE LOAD)

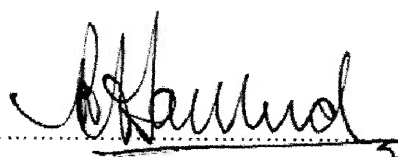
<u>ITEM</u>	<u>WEIGHT (kg)</u>	<u>LEVER ARM (mm)</u>	<u>CAPACITY</u>
Usable fuel (max)	110.0	3136	154.0 litres
Engine oil (inc in Basic Weight)	15.0	960	16.0 litres
SMOKE OIL	19.74	5080	21 LITRES
			<u>MOMENT (lb/in)</u>

Baggage: Refer to Flight Manual

NOTE: Fuel density is 0.72 kg / litre. Oil density is 0.94 kg / litre

To obtain the total loaded weight of the aircraft, add to the Basic Weight the weights of the variable and disposable load items to be carried for the particular role.

This schedule was prepared on **18-Sep-09** and supercedes all previous issues.

Signed.......... on behalf of Planeweights Ltd.
CAA Approval no. **AI/8538/79**
ISO 9001-2000 Certificate No. GB93/2024

NOTE: It is a requirement of the Air Navigation Order that the Commander of an aircraft registered in the United Kingdom shall satisfy himself, before takeoff, that the load carried is of such weight and is so distributed and secured, that it may be carried safely on the intended flight. Refer to the ANO, Article 28.



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G-CGFS

AIRCRAFT WEIGHT CONTROL SHEET

Use this form to record aircraft modifications which affect the weight and balance. Add/subtract the weight and moment of the item to/from the weight and moment of the aircraft. Divide the new aircraft moment by the new aircraft weight to obtain the arm (CG). Insert the new aircraft weight, arm and moment in the columns on the right.

		<u>WEIGHT (kg)</u>	<u>ARM (mm)</u>	<u>MOMENT (kg/mm)</u>
AIRCRAFT AS WEIGHED ON	18-Sep-09	1069	2580	2757907

Date and details of item added or removed